

IN THE CLAIMS

Please amend the claims as follows:

Claim 1 (Currently Amended): A turbine nozzle, comprising:

airfoils stacked along ~~the~~ a stacking axis,

wherein ~~the~~ high curvature portions on a suction surface in airfoil ~~section~~ sections successively formed along the stacking axis ~~of the airfoil~~ describe a parabola line that curves toward ~~the~~ a pressure side of the airfoil when seen from ~~the~~ a front or a rear of the turbine nozzle, wherein ~~the~~ a point of maximum high-curvature in each airfoil is located at a midpoint ~~portions on suction surface in airfoil section curve most at the center along the stacking axis of the airfoil from a straight line that connects with a first intersection between the parabola line and an inner band in the turbine nozzle and with a second intersection between the parabola line and an outer band in the turbine nozzle, and wherein a distance of the point of the maximum curvature falls to the straight line is within a range from 0.02 to 0.03 fold~~ 2 to 3 % of a distance between the first and second intersections along the stacking axis of the airfoil.

Claim 2 (New): A turbine nozzle, comprising:

an inner band;

an outer band; and

a plurality of airfoils disposed between the inner and outer bands, each airfoil of the plurality having a convex pressure side, a concave suction side, an X axis along a stacking direction, and a Y axis perpendicular to the X axis, the X axis being defined by a straight line connecting first and second points at intersections of the airfoil with the inner and outer bands, respectively, the first and second points defining portions of maximum airfoil

curvature on the concave suction side of airfoil cross sections perpendicular to the X axis, and the Y axis intersecting the X axis at a midpoint between the first and second points, wherein concave suction side profiles defined by cross sections parallel to X-Y planes are parabolic when seen from a front or a rear of the turbine nozzle, and

$$0.02 H \leq Y_{\max} \leq 0.03 H ,$$

where Y_{\max} is the Y coordinate of the midpoint and H is the distance along the X axis between the first and second points.

Claim 3 (New): The turbine nozzle according to claim 2, wherein an airfoil reduced velocity ratio for a lower-degree vibration associated with torsion is 0.7 or less.

Claim 4 (New): The turbine nozzle according to claim 3, wherein a ratio of a natural frequency of the airfoil to a natural frequency of an airfoil without curvature is greater than 1.0.

Claim 5 (New): A low pressure turbine of a gas turbine comprising the turbine nozzle according to Claim 2.

Claim 6 (New): A high pressure turbine of a gas turbine comprising the turbine nozzle according to Claim 2.

Claim 7 (New): A gas turbine engine comprising the turbine nozzle according to Claim 2.

Claim 8 (New): The turbine nozzle according to claim 1, wherein an airfoil reduced velocity ratio for a lower-degree vibration associated with torsion is 0.7 or less.

Claim 9 (New): The turbine nozzle according to claim 8, wherein a ratio of a natural frequency of the airfoil to a natural frequency of an airfoil without curvature is greater than 1.0.

Claim 10 (New): The gas turbine nozzle according to claim 1, wherein a trailing edge of each airfoil is parabolic.

Claim 11 (New): The gas turbine nozzle according to claim 2, wherein a trailing edge of each airfoil is parabolic.

Claim 12 (New): A low pressure turbine of a gas turbine comprising the turbine nozzle according to Claim 1.

Claim 13 (New): A high pressure turbine of a gas turbine comprising the turbine nozzle according to Claim 1.

Claim 14 (New): A gas turbine engine comprising the turbine nozzle according to Claim 1.

Claim 15 (New): A gas turbine engine having a turbine nozzle with redesigned vanes, the turbine nozzle being according to Claim 1.

Claim 16 (New): A gas turbine engine having a turbine nozzle with redesigned vanes, the turbine nozzle being according to Claim 2.